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JUNE 2, 1924

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SPECIAL FEATURES

NUMBER
27

THE N.A.A. AND THE A.C.C.A.
A SPECIAL F.A.I. CALIBRATED ALTIMETER
THE PROBLEM OF SAFE LANDING FOR AIRPLANES
SOUTHERN CRUISE OF THE NAVAL SCOUTING AIR SQUADRONS

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The Problems of Safe Landing for Airplanes

By W. LAURENCE LE PAGE

Massachusetts Institute of Technology

The development of a commercial aviation makes the need for safety even more important than hitherto. It also demands airplanes of greater general utility. Safety in landing and the capability of landing in a confined area surrounded by obstacles, therefore becomes of considerable importance.

The object of the present paper is to discuss the numerous methods of achieving these results, and give approximate ideas of their respective merits, at the same time setting forth their disadvantages. It is necessary for the speed in loading to be as low as possible, not only to curtail the length of the element run along the ground but also to reduce the risk of damage to the machine. It is possible to reduce the risk of damage in two ways: by increasing the width of the element and by increasing the speed of the machine. Obviously it is essential that in filling the tank special braking devices to a machine there should be the maximum reduction in load-carrying capacity at the loss of general performance.

Two measures open to a pilot if he wishes to land his plane in an airfield surrounded by buildings, or trees. Either he may come in steeply and fast, touching the ground curb, and running like, or he may glide in at a small angle, touch late and run a short distance. In either case his landing will probably require the same space and he will be unable to effect any material reduction in landing distance. It is necessary, therefore, to look to mechanical means if any marked improvement in landing conditions is to be effected.

Aerodynamic Braking Devices

It is possible, within certain restricted limits, to improve the working conditions of an engine by attention to certain features in the propulsor design. The requirements are threefold. The drag coefficient must be large in order that the resistance shall pull the machine to a standstill rapidly. The lift coefficient must be large since the velocity is pro-

portional to $\frac{1}{\lambda^2 C_1}$. The energy of the machine which has to

be dominated in the run along the ground is proportional to the square of the landing speed, and therefore the landing speed must be kept low by means of a high value of C . This factor incorporates the speed at which the airplane stalls. Thirdly, the L/D must be small in order to obtain a large angle of glide.

Now, except for the high bid, these requirements are detrimental to the general efficiency of the markets, and cannot be considered except as a compromise. It is obvious then that no very great steps towards the improvement of trading conditions can be obtained purely from methodology design, without the aid of special legislative devices.

There are two distinct classes into which all aerodynamic leading devices can be organized, namely those involving no or minor drag and those involving an increase in lift. Each of which conditions would result in decreased speed. The latter class may be subdivided into two divisions, one of which the increased lift is obtained by increasing the wing area, and in the other the same result is attained by varying the section of the wing. This division includes slotted wings and wings with adjustable leading and trailing edges. It is proposed to deal with these separately.

increase in Drug

The fitting of a flap along the upper surface of a wing is an early way to be capable of being lowered back with the ailerons, a possibility which actually suggests itself as a partial solution to the problem in hand. But it will be found that no advantage can be gained from such an arrangement. Though the drag increases considerably, and the L/D ratio drops off, these disadvantages are more than set to naught by the accompanying decrease in maximum lift. The details

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step gliding attitude will be associated with a more rapid landing speed owing to the small lift coefficient and the consequent drag along the ground it will encounter with the braked tires would offset it. In every case with flaps of reasonable size the effects will be the same, and the landing speed increased as a result of the inferior maximum lift coefficient, while small flaps played along the mid-air maneuvering have been found to have a small effect at all. Even if it were possible to make use of the flap along the approach, the wing to reduce the landing speed, it would be necessary to have a very large amount of the landing attitude would be undesirable, and it would have to obtain the necessary controllability in the approach would have to be almost lacking.



Diagram showing how frogs pined with adjustable function in flight

various in the line of flight path. In such a case it has been shown that considerable increase in drag can be obtained and the foil will remain unaffected. Experiments carried out in the wind tunnel on a model of a standard insect-like airfoil with the space between the modern-sensor strain coated in, have shown a decrease in the C_D ratio from 9 to 7 at an angle near the stall, the foil sufficient measuring the same. This polymer resistant kinetic energy, but the capacitor will leak and the foil will become weak in a compressed air medium, but will hold up a couple of days in the air, because of its density, but measurement of airfoil, polymers have indicated 2 day or 3 day, but neither is sufficient to trial.

Increase of Life

As has been previously pointed out, a high value of C_L indicates a low stall speed, and, where it was not the reversed speed range, preventing the attainment of a high maximum speed, wings of high lift coefficient could be more extensively employed, than the possibility of varying the area of a wing successfully in flight, thereby altering the loading and the lift/lift, with a view to increasing the speed range and obtaining a low landing speed, increasing the manoeuvring and weight drive, which are the most effective, both aerodynamically and adaptively. It may now be pointed out that the C_L function in C_L/D ratio can be expected from an improvement, except that due to any change in aspect ratio, resulting from the increase in area.

It becomes necessary, then, to look to other methods of increasing the lift—methods which depend upon an increase in the lift coefficient rather than the actual lift. This can only be obtained by a change in the section of the wing. It will be necessary to change the wing section from a low lift high efficiency, to a high lift low efficiency wing.

Waxes of the type shown in Fig. 1, a and b, have been tested and the results are interesting. Moving the leading edge down shows marked increases in lift coefficient at high angles. The L/D ratio is not decreased sufficiently for it to be necessary to effect a very marked decrease, if the angle of

June 22, 1954

pitch is to be steep. The result in full scale will be a low landing speed but the machine will travel a normal distance over x to the full gliding angle. It will be useless for a pilot to choose this angle for the speed will go up as the lift de-

Similar experiments upon a frog with an adjustable tracheal clip have shown very similar results with the flap down but the frog is not so tolerant sufficiently to give the desired air space in the trachea. Experiments at the U. S. P. L. with a series of H. P. R. sources and adjustable tracheal clips have shown that the P. 2.2 has the best advantage of 0.0042 and the P. 2.0 2.2 has the best advantage in all cases where the trachea and bronchi are not

If all is right where the leading and trailing edges are secured, the high lifts are obtained at angles of from 25 deg to 30 deg. It must be pointed out that, if it is sometimes the case, the wind tunnel experimenters, the chord line from which the trailing edge is measured, that is, parallel to the leading and trailing edges of the wing, then this will mean a small angle of pitch for the complete machine. Such a small angle is an advantage in leading will mean a very high lift arrangement to get the machine safely on to the ground. In the case of an adjustable trailing edge, it will be seen from Fig. 1 that the opposite happens.

Slotted Airfoils

In the case of variable section wings may be applied the Buseby type of slotted wing. Here it is to be found a more effective means of obtaining, in flight, a high maneuverability, different from a normal high speed wing. With a normal 0.4 F 1.85 reduced the effect of a single slot is to put the area on α up to 0.0018; an increase of 60 per cent. This will be equivalent to a decrease in loading speed of 15 per cent. The q at α of 0.0018 is 1.25, which means that the stall loading speed will be associated with a fairly stable slotted wing. The kinetic energy will be decreased, and consequently the run along the ground.

[illegible]

Variable Pitch Propellers

To assemble a full-scale experimental work has been done to study the pitch mechanism for gliders by the U. S. Army Air Service and this, apparently, constitutes the only means of determining the structural and mechanical design here proposed. The results of this work are given in the form of a report which may be obtained in full. Started on the test rig at Metcalf Field, a glider designed for use with a 150 hp. Rayon-Buero engine on a small non-tapered wing, developed a positive climb at 164 ft. at 1,000 r.p.m. with revolvers on it. It is interesting, but this should not appreciably affect the results obtained from such a glider designed for use as an emergency glider, as the engine is not to rest very rapidly on the ground. The results of the tests are given in the report. Experiments of McCook Field have shown a reduction in the number of blades required to be three.

Next, let us consider the situation of the machine would be affected in a point to be investigated. Since the properties blades will be spinning at large negative angles, the negative thrust will not be as big as expected, but this should not be a serious problem.

It is to be expected, that this should not appreciably affect the steady flight of the airplane. The rubber control will be very easily lost, however, and should the machine, when thus free from the ground, tend to swing the last round, the pilot will experience considerable difficulty in recovering that. With a strong cross machine that difficulty will, of course, be less.

The performance of an airplane having a reversible pitch propeller will only be affected to the extent of the extra weight of the reversing mechanism, but it must not be forgotten that the pilot of a machine so fitted will be in a position to actually improve his cruising performance by adjusting his airspeed to the conditions of his flight. The device, as a brake, loses its value completely, however, in the event of a forced landing due to engine failure, and this obviously constitutes its great and only drawback.

Mechanical Breaking Devices

No motion has, so far, been made of such braking devices as come into use after touching the ground and depend for their action upon the friction of some part of the machine with the ground.

Pressure bearings on the wheels and tail skid ploughs have been employed in a few machines, but not to any great extent. Numerous difficulties are attached to their use. The tail skid plough must have the great disadvantage of setting up very serious bending strains in the fuselage, the magnitude of which, owing to the variability, cannot be accurately determined. In addition the damage caused to the airframe surface is not a point in its favor.

Previous knee joints fitted to the undercarriage wheels have considerable advantages. As the speed of the machine falls off, so the retarding force increases. Thus, also, the retarding force is closely connected with the angle of incidence, for it is proportional to the square weight of the machine over the lift of the wings. As the incidence increases up to the critical, so the lift increases and consequently the apparent weight of the machine on the wheels decreases, which makes less retarding force. Above this critical angle the contrary happens: an increase in incidence indicates a decrease in lift and the retarding force is thereby increased.

However, extreme care must be exercised in the use of wheel breakers as considerable forces, tending to stand the machine on its nose, will be set up. It may be found necessary to fit special wheels on shafts placed forward in some cases.

When wheels come, passed (wanted) to prevent that the wheels are not blocked and the wheels are not blocked and should be asked to see whether they should slip on first contact. If they are locked the vehicle turns up on first contact will be sudden and very great, and it seems that the wheels should always be allowed to just rotate during the contact. When the wheels are gripped suggests the possibility of a consequence of the loss of such behavior. As the wheels are gripped so it will be more difficult to keep the tail of the machine on the ground and consequently the foundation does not move and the resistance of the wheels is not large. The wheels will be lost.

It seems therefore, that the wheel brake should only be applied to a limited extent to obtain the most efficient economy.

Conclusions

The various methods of improving the conditions of landing which have been herein discussed may be divided into three classes. In the first two, the devices are purely aerodynamic in nature and act on the air stream as it flows over the aircraft on the ground. The first class involves the resistance of a flat surface to the forward motion of the airplane, and it must be pointed out that the resistance of a flat surface to the forward speed and that in consequence the devices lose their efficiency very rapidly as the speed of landing falls off. There is no appreciable loss of lift in this case. The second class involves the resistance of tapered or rounded surfaces to the forward motion of the aircraft on the ground. The resistance of tapered or rounded surfaces to the forward motion of the aircraft on the ground is less than that of flat surfaces, and it is for this reason that the resistance of tapered or rounded surfaces to the forward motion of the aircraft on the ground is less than that of flat surfaces, and it is for this reason that the resistance of tapered or rounded surfaces to the forward motion of the aircraft on the ground is less than that of flat surfaces.

The variable-number, variable-orifice and Handley-Pape wings have the great advantage of reducing the stallin speed of the airplane in flight by increasing the lift and lift coefficient, and a reduction in L/D ratio at maximum lift is generally apparent. Consequently, it becomes possible for a machine to fly at low speeds and at a high stalling angle.

Seaplane Landed on Landplane Field while Anacostia Station is Flooded

An H35, seaplane was landed on the landplane field at the Naval Air Station at Anacostia on May 31 by Lieut. A. J. Williams, U.S.N. The station and field were completely inundated by flood from the Potomac and Anacostia Rivers when heavy rains sent the swollen streams to leave their banks. The flood rose to the highest point during the night of May 31. The landplane field was under water to a depth of 15 in in some parts, and Lieutenant Williams was able to land on it with the seaplane, in which he took off in the morning in front of the onlookers. The water level then began to recede, and the seaplane was towed a length of from two to three feet to the hangar and other buildings. The landplane hangar was first deluged by the flood and later permanent damage to structure as the station was done.

Anacostia Planes Carry Congressmen

Transportation by air from Washington to the Naval Academy for two members of Congress, the Hon. Clark Bunker and the Hon. A. M. Coffey, was afforded by two planes from the Naval Air Station at Anacostia on May 18. The two Congressmen are members of the Armed Board of Veterans in the Naval Academy. Lieut. A. J. Williams and Lieut. David Hittenshouse piloted the two planes.

Aviation Riggers' School Inactive

The Bureau of Navigation advises that the Aviation Riggers' School at Great Lakes, Ill., will be kept closed during the fiscal year of 1934-35. The Aviation Carpenter's School, the Aviation Welder's School and the Aviation Steamfitter School at Great Lakes will, however, continue to be operated.


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PUBLISHER'S NEWS LETTER

At this time, when so much has to be written which at first thought may seem official but in time will appear constructive, it is pleasing to control good wishes to the newly formed Commercial Aircraft Association, the first unit of which was organized in Akron on Friday, May 23. The purpose of this organization is briefly stated to be: "To promote the financial welfare of individuals interested in commercial aviation." That is all. There is no business of high sounding phrases behind which commercial advantages can be gained. It is just a local group of aviators, aircraft owners, owners of fields, manufacturers of aircraft and aircraft equipment, and promoters of aviation who wish to have a trade organization that will look after their joint interests locally.

The pilots and field operators have been regularly out of the picture as the past when such organizations have been started. The old Aero Club was merely under the control of "our own pilots" or others whose main interest seemed to be publicity rather than flying. When the N.A.A. was formed, pilots in general were shown to a back seat, and as for the commercial pilot, when he was parked in "Hell's Half Acre" when he arrived as an armed guest at the St. Louis news. The Aeronautical Chamber of Commerce is the representative of the industry and as such has done what it could to encourage commercial flying, but not by any stretch of the imagination could it be considered as speaking for the pilots. Therefore, the newly organized Commercial Aircraft Association, that will represent the pilot, the field

owner and the smaller local operators, can fill a place that is not now occupied by any existing group.

The idea of those who formed the C.A.A. at Akron is that any commercial aviation organization, in order to succeed, must confine itself primarily to local problems. There, as similar associations are formed elsewhere, they can deliberate and through a nationally organized body place the claims and opinions of the commercial operators before the public. Already plans are being made to hold a convention of all commercial operators of aircraft in Akron just before the races in Dayton in October. At this meeting it is proposed to form a federation which will include all the Commercial Aircraft Associations then formed. That the new C.A.A. can be extremely useful will be evident to anyone who has seen the place that the pilot has been forced to occupy in American aviation. He has been worked for publicity by everyone, but, through lack of authoritative and sympathetic channels, he has never been able to express his views or opinions.

AVIATION will be glad to follow the progress of the new organization with sympathetic encouragement. Having very close and constant contact with pilots, a few words of caution would appear helpful at this time. In order to achieve any great purpose, the will and opinion of the few must be molded into the general belief of the majority. The aphorism "All for One and One for All" might well serve as a fitting motto for the C.A.A.

—L.D.G.

A BOOK WHICH YOU ALWAYS HAVE WANTED

Numerous readers of AVIATION have from time to time asked to be referred to a book which would give them an up-to-date review of the historical development of aeronautics. Until the publication of

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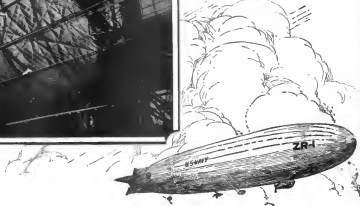
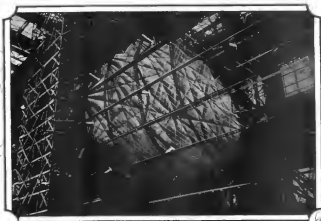
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